

# Design and Development of Ejection Seat Mechanism in Aircraft



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**Abstract:** This research manages the discharge framework that can be actualized in a common air ship. It is significant for a flying machine to have a launch situate in the event that the plane meets a mishap in a fight or during experimental drill and the pilot needs to rescue to spare their life. The component includes lifting of the pilot to a protected good ways from the air ship and after that sending the parachute. As there is increment in air traffic, this discharge framework gives extra security measures to the life of travelers. By this strategy alongside the pilot, the group individuals and the travelers can be spared. This discharge framework comprises of an exceptionally made ejector cushion, track framework and parachute. The launch framework is administered by a planning sensor which guarantees that the development of the traveler seat occur in equivalent interim of time to maintain a strategic distance from impact between two continuous seats.

**Keywords :** Bail, Ejector pad; Track system; Ejection; Hydraulic; Propulsion; Stability.

## I. INTRODUCTION

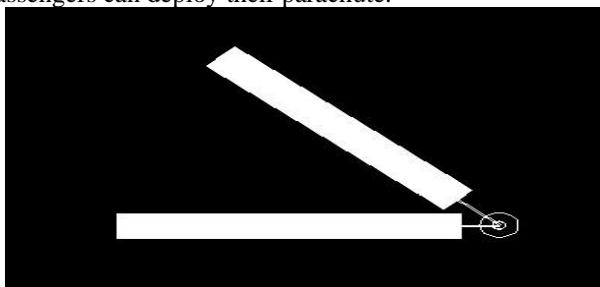
Ejection seats have been customarily fitted in military air ship from the late 1940's and onwards. Military pilots wore appropriate suit which can be a bit of leeway lifeline if the discharge needs to impact at high heights, however was extremely troublesome and delay in sprucing up and checking activity of wear with travelers. For the powerful discharge of the considerable number of travelers in an enormous non military personnel air ship it would need to be incorporated to break down with predefined parts allowing everybody inside a reasonable possibility of getting security. The deformity in a manner as once the fuselage is separated in generally or in a mishap the inhabitants have extremely poor shot of survival. The launch seats weigh extremely high and when inhabitants catapulted and conveying with various parachutes causes crash. What's more, it increment weight

which isn't appropriate to fly an air ship. For moving of seats it need high fuel and cost of such carriers would be essentially astronomic. In the contender flying machines at the hour of launch the shade is first opened and the discharge seat is rescued with the pilot, this framework can be additionally utilized in the common plane to rescue the pilot, yet as by the common airplane laws, the pilot is in charge of the security of the travelers. Simultaneously the travelers can't be catapulted like in the contender planes since we can't open the top of the plane and can't discharge huge number of travelers one after another. To do so we need high propulsive capacity to launch every one of the seats. On the off chance that we do so it might cause impact of the progressive seats that launched out and it made major issue. For this it need Catapult, Rocket, Restraints and Parachute for every one of the travelers. It will expand the heaviness of the flying machine. Increment in weight isn't great to fly a plane. To conquer these issues of increment in weight, need of high propulsive power and seat crash the technique here included has less increment in weight which only more than void weight. Rather than high propulsive power, straightforward spring system is actualized. To stay away from the impacts the planning framework, succession and equivalent interim of launch is incorporated. In this strategy there is no compelling reason to open the top of the plane just we can open the side entryways and crisis exit to launch the seats.

## II. METHODS AND DESIGN

### A. Ejector Pad

The ejector pad is specially designed to move with a concept of 'constant angle varying speed'. The ejector pad is placed at the exit doors. Instead of ejecting the seats over the top, the seats are ejected in side ways to some distance and the passengers can deploy their parachute.



**Fig 1. Design of ejector pad**

The ejector pad is constructed with a high power mechanical spring system and hydraulic pistons which produce ejecting power to the pad instead of providing propulsion power to the seats separately. It reduces the propulsive power for each and every seat.



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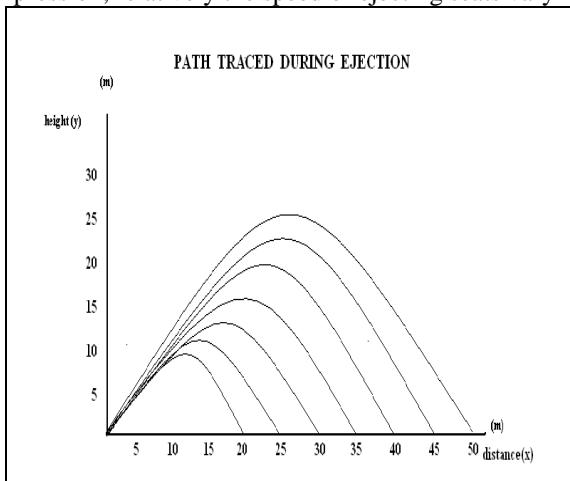
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There is an only common ejector pad at every exit. If the piston in the hydraulic system which is placed under the ejector pad is increases the compression of the spring increases, which increases the ejecting power. There is a releasing lever at every pad. It holds the ejector pad during the compressing of the springs. For every seat the spring gets compressed and the releasing lever is opened and the seat gets ejected. The pistons are held over one another. For successive seats the piston height is increased mean while the tension of the spring will increased it will made the seats to ejected with different pressure and it acquire different distances.

The procedure is repeated for all the respective rows and seats. With the successive increase of piston and compression, relatively the speed of ejecting seats varying.

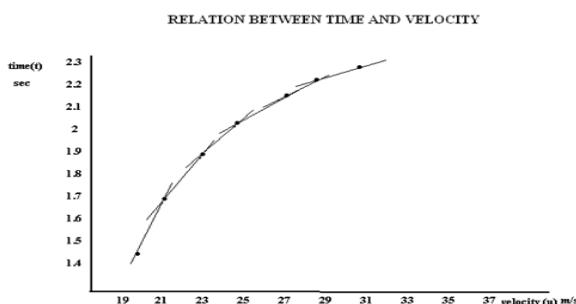


**Fig 2 path traced during ejection.**

**Table 1. Height and distance of path trace**

HEIGHT (y)	10	11	13.31	15.77	18.47	21.38
DISTANCE (x)	20	25	30	35	40	45

This is the height and distance travelled by successive seats. By this procedure the collisions of the seats get avoided.



**Fig 3 relation between time and velocity**

In this the each line drawn between the points denotes that each seat has its own path, time and velocity.

**Table 2. Relation between time and velocity**

Time(t) sec	1.42	1.68	1.84	1.97	2.09	2.19	2.28
Velocity (m/sec)	19.8	21	23	25	27	29	31

The varying speed will avoid the collision of the seats. This varying speed system is sited different in other exits by in ascending and descending of speed.

Consider the ejected seat travelled in a parabolic path the time interval ( $t$ ), the height projected( $y$ ), and distance projected( $x$ ) can be calculated with the formula.

$$T=x/u\cos\alpha$$

$$Y=u \sin\alpha t - \frac{1}{2}gt^2$$

Where in the ejector pad moves at an angle of ( $\alpha=45^\circ$ )

For velocity of  $u = 19.81 \text{ m/s}$

The height projected ( $y = 10 \text{ m}$ )

The distance projected ( $x = 20 \text{ m}$ )

And the time taken = 1.427 sec.

## A. Design of seat and wear:

The seats here employed are similar to the ordinary seats in the aircraft. A simple modification is made in the seats. A ball type bush is placed at bottom stand of the seats it is free to move. At the base of the floor the tracks are designed this is like the rail tracks but placed horizontally. There are gaps provided in the tracks to hold the seats.

The bottom bush of every seat is placed in the gaps between the tracks. The seat cannot be released unless the gaps get elaborated.

Additionally the suitable wear for the ejection is attached with the seats it can be closed with a belt. It covers the passenger as a bag. This bag like cover is attached to the seat with pressure buttons. When the seat is ejected out of the aircraft during the parachute get open these pressure buttons get released and the passenger separated from the seat. At the back of the seat the drogue parachute and a parachute and oxygen pouch is placed.

The opening of the parachute is in the hands of the passenger and it is also connected to a barometric system. The system is ejected along with the seats it calculates the height at which it flies. The height which is suitable for the deployment of the parachute is feeded. Each seat having a sensor unit. When the barometric box read the height suitable for deployment of parachute it send signal to the sensor then the parachute get opened. We cannot teach every passenger how and when to open the parachute and it is impossible. And at the time of ejection nobody was in normal condition to open the parachute at the correct height. These parachutes are designed like the gliders that it tends to the easy landing.

Hence there is no work and confusion for the passengers it is very simple to do such operations.

## B. Track design

The track is designed like the belt in the tanks. It moves around a cylindrical runner which is connected to a motor. This track provides small ball shaped gap between the pieces. This gap provides space for the seats to fix. The track is able to move around the runner. When the gap reaches the runner it get elaborated and due to limiting friction the seat released from the pad and moved to the ejector pad.

This is the side view of the track system the place at which the seat released to ejector pad is noted.

In this track design the right side first row and left side first row only able to move but other rows are fixed. And this rows act as the pavement. When the first row starts moving the seats in the row get start their ejection. After the first row the second row is moved laterally. And replace the place of the first row.



This lateral movement is made by the simple hydraulic system. This system moves the corresponding row after finishing of ejection of rows.

There is small run way for each seats for which the lateral movement is provided between the tracks. When the piston moves the row the seats moved laterally in the tracks then it reached the moving row. After that it gets into the runner and the seats get ejected.

#### C. Ejector system:

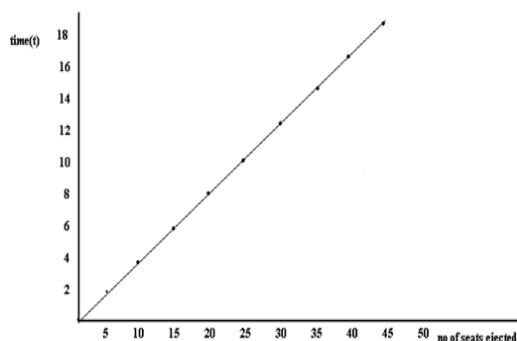
The ejector system is designed perfectly, and provided with perfect timing sense. This ejector system overcomes the disabilities of the implementation of ejection seat in civil aircraft. For a civil aircraft the problems are arrived in the form of repair in engine, birds hit and may some other problems in the structural and maintenance problems.

But there is no such a situation like a fighter aircraft face that a missile will attack it. So a civil aircraft may have a minimum 30sec of time for the complete disintegration. Thus this design runs at a maximum time limit of 25 to 30 sec only. The major problems that are considered as the weight, propulsive power, cost, mode of application and difficulties. In this design there is no need to open the top of the aircraft to eject and no propulsion is used to propel the seats thus it reduce the cost and fuel consumption and less usage of power.

Here we are not using the ejection seats used in fighter aircrafts. We are using only the ordinary seats used in the civil aircrafts with some additional equipment like passengers wear, parachute and a small oxygen pouch for the convenient breath of the passengers. This ejection system provides convenient ejection to the passengers rather than that of other ejection system in fighter aircraft. The power for the running motors and the track system are derived from the aircraft power system.

#### D. Timing sector:

Timing plays an important role in this system. Each seat in the track moves in perfect time sequence. Each seat in the track ejected with an equal interval of time. Within two seconds five seats can be ejected in this system. Each seat accrues its own time of flight, path, speed and distance ejecting. This term avoids the collision of successive seats.



**Fig 6. Relation between time and velocity**

This straight light indicates that the seats ejected with equal intervals of time.

**Table 3. Time taken for seats to eject**

TIME (t)	2	3	6	8	10	12	14	16
NUMBER OF SEATS	5	10	15	20	25	30	35	40

#### Operating System

When the aircraft is at a situation, there is no way to survive then only the ejection system will be operated. All the systems are not only controlled by the pilot but also by the commands from the ground station. At the situation to eject it cannot be done by pilot, he have to get the command signal to operate the system from the ground station. If both the pilot and the ground station commands are executed then only the system get starts.

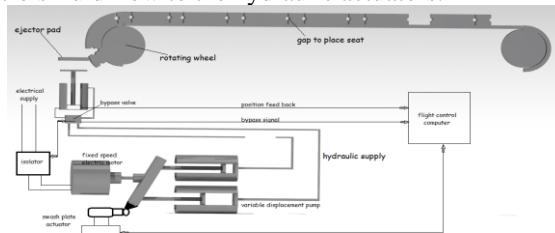
In spite of the situation the aircraft got away from the ground control system it disconnects its commanding suggestions. Then the operating system is free from the ground station and the full control is only in the hands of pilot.

This type of operating system is to be included because for the following situation. When in the case of the aircraft is hijacked by someone if the system has the control only in the aircraft. Then there is a chance of operating the ejection system without the knowledge of pilot. And it create serious problem.

For this situation the system must be controlled by both ground and pilot. And they can have their own possibility to operate the system when the system is disconnected from any one commanding prompt.

#### E. Hydraulic system:

This is an electro hydrostatic machine with dual channel. Essentially it contains elements of motor, pump, by pass valve and piston jack. It is uses a unidirectional fixed speed electric motor to drive a variable displacement pump that controls fluid flow to the hydraulic actuators.



**Fig 7. Hydraulic system**

With the pump rotating at constant speed a separate control mechanism varies the swash plate angle with pump and is therefore able to continuously vary the hydraulic fluid flow. When the swash plate passes through the over centre position, hydraulic fluid flow is reversed. Thus it is changing the direction.

It not needs high power electronic controllers. AC induction motor connected directly to the aircraft AC supply

- This system is designed less sensitive to loads.
- And it has rapid start up response.

The swash plate is connected to an actuator which controls the (angle) movement of the swash plate. By increasing the angle the fluid flow will be more at one condition. Thus it increases the lifting of piston in hydraulic. Where it varies the tension of spring for the successive and varying speed of the seats to be ejected.

All these hydraulic cylinders are connected to a single controlling system where us it reduces weight and it is convenient. Then we can simply place the hydraulic under the ejector pad.



The isolator in the system controls both the electric motor and the bypass valve. Whereas an electrically controlled system is always easy to operate and the electric power can be stored. Isolator controls the speed of the motor and the flow of fluid in the bypass valve. An additional setup called position feedback is derived from hydraulic piston. This can be used to find out the position of the piston.

The major connectivity to all these systems is the flight control computer. Whereas the position feedback and bypass signal and swash plate actuator signals are feeded and these can be controlled by the programming feeded in the system.

### III. RESULTS AND DISCUSSION

Pressure and temperature plays an important role in the flight. The pressure at the ground and at certain altitude is varies. Therefore the aircraft acts as a pressure vessel. Due to this reason the aircraft is pressurized. If it not pressurized the aircraft will burst due to pressure variation at inside and outside. So the pressure is maintained inside the bulkheads are made in such a manner that to expand at high altitudes and it will come back to normal at low altitudes.

The temperature and pressure variations are calculated using the hydrostatic equation.

$$\frac{dp}{dh} = - \rho g$$

The pressure and temperature varies with altitude.

Altitude hg (m)	Temperature (K)	Pressure N/m <sup>2</sup>	Density kg/m <sup>3</sup>
0	288.16	$1.01325 \times 10^5$	1.2250
1000	281.66	$0.89876 \times 10^5$	1.1117
5000	255.69	$0.540448 \times 10^5$	0.73643
10000	223.26	$0.26500 \times 10^5$	0.41351

There is problem arise that we cannot open the aircraft at certain altitude. Explanation like that an aircraft is like a balloon in will be burst when there is a pressure variation at inside and outside atmosphere. The same concept is included in the aircraft.

Let us implement this idea that at sea level altitude say 0m the temperature, pressure and the density is tabulated. At this stage the balloon is just filled with air, here the condition is same at inside and outside of the balloon is same. If we hit it with a pin it does not bust.

Like this the aircraft will not bust at ground level when there is open in it. Consider at an altitude of 1000m the pressure in aircraft is like the balloon filled with 20-30% of air here also there is no possible of bursting. At 10000m it is like 50-60% and at 15000m like 65-70% of pressure so if we hit it with a pin the air inside comes out but it does not burst likewise the aircraft will not burst at these altitudes but there is variation in the stability. Hence this ejection system will be operated at altitude below 15000meters.

You can ask what about for the above altitudes?

For an altitude more than 15000m the pressure variation is high and if we bailed out it will create serious problems to the passengers. When there is a problem in that altitudes the ejection is not possible. If there is failure in engine or some other problems the aircraft will descent because of need of thrust and high drag. So when the aircraft comes into the altitude below 15000m we can operate the ejection system.

#### Stability During Ejection

The stability of the aircraft will changes when we open the aircraft at certain altitudes. The stability and control is focused on moments on the airplane and the moment on the control surface.

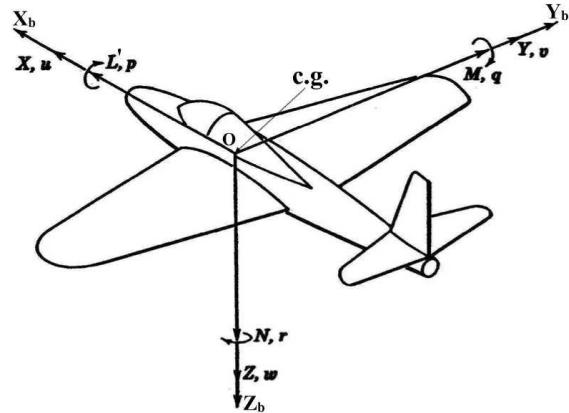


Fig 8. Moments of aircraft

Rotational motion about the x, y and z axis is called roll, pitch and yaw and the L', M and N are rolling, pitching and yawing moment respectively.

Consider a commercial airplane flying at a altitude of 1000m at this altitude the emergency exits are opened for the ejection at this situation the airplane tends to have stability variation at X, Y and Z axis that is about rolling, pitching and yawing moment.

In the rotation moment at in stability condition there is a possibility of the airplane will rotate at any one say left. At this situation if we bailed out the passengers it would not made them collide with each other because rotation about one axis tends it to the centrifugal force which made them through away from the airplane. Same way in pitching the aircraft can pitch up or pitch down at this case seat ejected at front exit reaches a height greater than normal height of ejection because of pitching where the nose is at pitch up. And the seat ejected at rear exit acquires less height than the normal ejection where the tail portion is down and this is vice versa at the pitch down condition. On considering the yawing moment the stability variation not much more considered during ejection.

### IV. CONCLUSION

As there is increase in air traffic the ejection seats in civil aircraft will play a major role in the safety of passengers. Simple design and perfect timing sense will made this ejection system a valuable one. Surely it will be a mile stone in the development of the ejection seat. This system can be implemented in big commercial aircrafts.

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## Design and Development of Ejection Seat Mechanism in Aircraft



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