

Research and Design of Submerged Vehicle's Propeller Blade

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Abstract— Recently, Fiber Reinforced Composite is used for making a propeller blade to develop its performance by increasing the payload and underwater speed of the vehicle. As a consequence of its feathery weight & high strength, numerous scholars/scientist substituted the conventional metallic material with composite material for crafting the propeller. In the contemporaneous work, predictions of pressure circulation around the profile of a propeller blade as a result of hydrostatic pressure difference are existing. Static structural stress investigation was executed for a single combination i.e. carbon fiber reinforced plastic (CFRP) & for hybrid condensation (a combination of two composite materials) i.e. CFRP & Glass Fiber Reinforced Plastic (GFRP). ANSYS APDL software is used to conclude von Mises pressure developed in the propeller blade. The result has been given away that it is more beneficial to use the hybrid blended material than the composite one. The weight value of propeller blade is found to be lower for the hybrid complexes, proposing the intrinsic worth of the contemporaneous work.

Keywords: Propeller blade, (CFRP), (GFRP), hydrostatic pressure difference, hybrid composite

I. INTRODUCTION

In this progressive era of science & technology, using our non-traditional resources as a source of energy has become an important issue. One of these important areas is an underwater maritime platform where much research effort has been done to develop underwater vehicles and to look at its different impacts. A propeller is one of the vital parts of an underwater vehicle for a propulsion/forward motion system [1]–[4].

The strong point of the propeller blade is lone of the most essential factors for the appropriate enactment of the underwater vehicle. Propeller blade is usually made of conventional metabolic elements like as aluminum alloy, stainless steel etc. Now a day due to light weight & prolonged durability, composite materials are used instead of metallic component [5]–[8].

In the past literature, Rao and Reddy concluded that composite propeller blades are safer in case of resonance onions in their harmonic analysis. Vibration defect can also be controlled in case of dampness effect any more [9], [10].

Khan et al. observed shear stress b/w laminar for composite materials considering different number of layers and found that the layers are strongly correlated. Eigen value investigation illustrates that composite material is 80.5% more natural in nature than aluminum propeller [11].

Ganesh et al. have carried out static and modal analysis for aluminum propeller and propeller (carbon-reinforced plastic). From the analysis, it is witnessed that the blade

definition in the case of a blended propeller is not significantly smaller compared to aluminum [12].

Yeo et al. predict stress distribution around a propeller blade by FEA technique. Considering a three laminated propeller and stainless steel as the material, a hydrodynamic study is approved out. By means of rising RPM/rotational speed, the modified pressure is also increased after 3000 rpm, exceeding its critical pressure [13], [14].

II. MODELLING OF PROBLEM BLADES

CATIA V5 application is employed for modeling the propeller blade. The shape design module is employed to generate a 3D model of propeller blade. The model is created by taking into account external diameter of propeller blade as 200mm and the inner diameter (i.e. diameter of propeller) as 30mm. The blade corners get a round shape with a radius of 40mm. The thickness of the applied blade for the purpose of analysis is 1 mm. A 3-D replica of the propeller vane is shown in *Figure 1*.

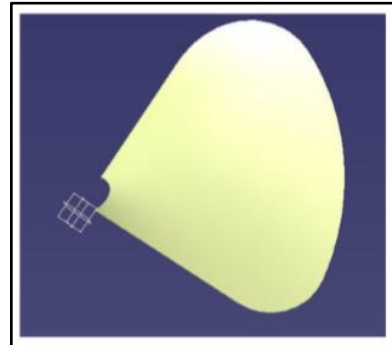


Figure 1. 3-D replica of propeller blade.

Mesh Generation

The meshed propeller vane/blade is displayed in *Figure 2*. The element used for meshing is 4 element 141 shell. Considering 2 mm element size created in preprocessor meshing module of ANSYS APDL [11], [15].

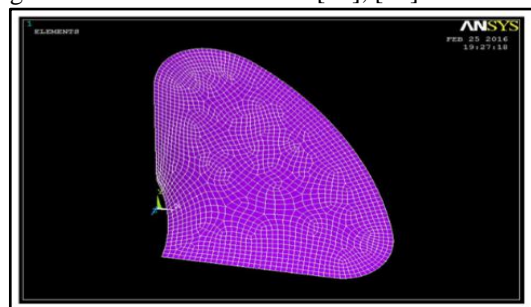


Figure 2. Meshed propeller blade

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Boundary Condition

The circumference of the blade that meets the center point is fixed for each translation level of freedom. During blade propeller analysis considered as cantilever beam. Loads are applied approximately 1/3 distance from blade tip i.e. at 0.7R to 0.75R, wherever R is propeller blade radius. The applied load size is equal to 4000N for static structural analysis and loads are distributed evenly among selected and normal nodes to the blade face.

Material Characteristics of Composites

The composites that are given in the table are described with their physical properties. The physical properties of the composites are presented in *Table 1*.

Composite material	CFRP	GFRP
Young's modulus E_X (Gpa)	22.925	120
Young's modulus E_Y (Gpa)	22.925	10
Young's modulus E_Z (Gpa)	12.4	10
Poisson's ratio ν_{XY}	.12	.16
Poisson's ratio ν_{YZ}	.2	.2
Poisson's ratio ν_{ZX}	.2	.16
Rigidity modulus G_{XY} (Gpa)	4.7	5.2
Rigidity modulus G_{YZ} (Gpa)	4.2	3.8
Rigidity modulus G_{ZX} (Gpa)	4.2	6
Density(ρ)(gm/cc)	1.8	1.6

Table 1. Physical Characteristics of materials

III. RESULT AND DISCUSSION

During analysis, various number of layers such as 4 rows, 8 rows, 12 rows and 16 layers are considered for both homogeneous and hybrid products. The thickness of each dressing cover is considered. In the case of hybrid coupling the amount of CFRP and GFRP fluctuates respectively as illustrated in *Figure 4*. Blended fiber orientation is given as 45° and -45° respectively as shown in *Figure 3 & Figure 4*.

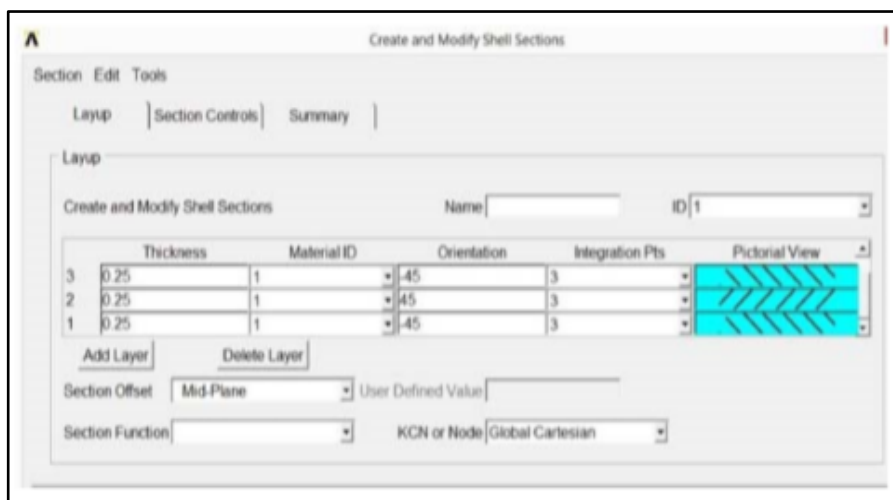


Figure 3. Various layer & fibre positioning of CFRP (4 layers)



Figure 4. Various layer and orientation of hybrid composite (4 layers)

Static Structural Analysis

Structural analysis is performed for four cases for both single CFRP and hybrid compound propeller considering various layers and corresponding illumination and stress are observed.

Case1: analysis result for 4 layers

The maximum direction for a single CFRP propeller blade is 0.802mm in a z direction that is, customary to the blade front & the maximum von Mises pressure is 22.1 N / mm². In the case of a hybrid propeller mounting shaft at 0.9213mm & maximum von Mises pressure is 21.1 N / mm². Both illumination & von Mises stresses are presented in Figure 5, Figure 6, Figure 7, & Figure 8.

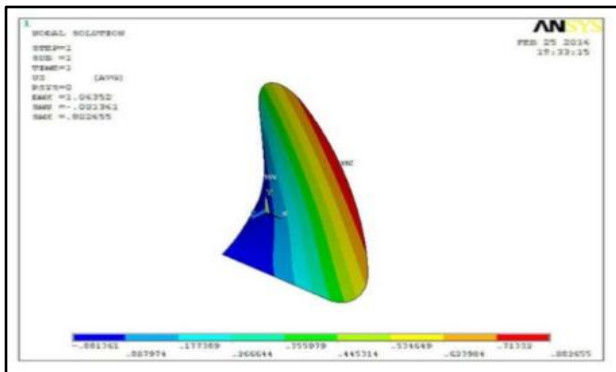


Figure 5. Deflection of CFRP composite blade (4 layers)

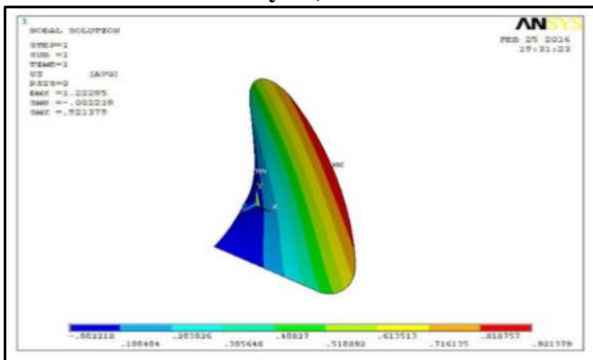


Figure 6. Deflection of hybrid compound blade (4 layers)

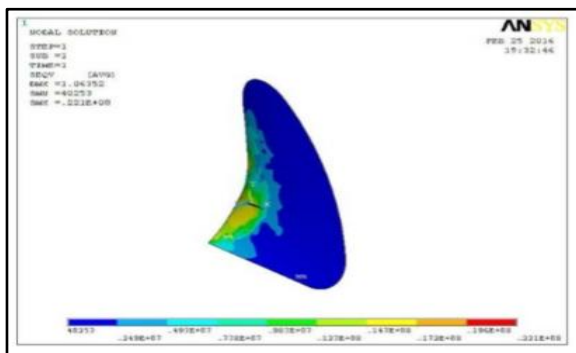


Figure 7. Max von/misses stress for CFRP propeller blade (4 layers)

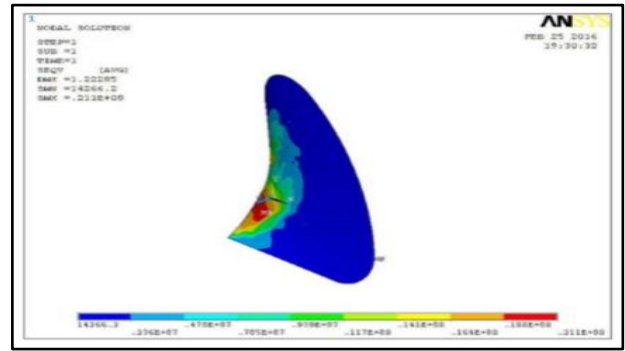


Figure 8. Max von Mises stress for hybrid composite propeller blade (4 layers)

Case 2: Analysis result for 8 coatings

Here the thickness of each cover is estimated to be 0.125 mm. The maximum illumination for a single CFRP propeller blade is 0.75 mm in z direction i.e. normal to the blade face & max von Mises pressure is 23.7 N / mm². In the case of a hybrid propeller shaft at 0.79 mm and maximum von Mises pressure is 20.1N / mm². Both illumination and von Mises stresses/pressure are displayed in Figure 9, Figure 10, Figure 11, &

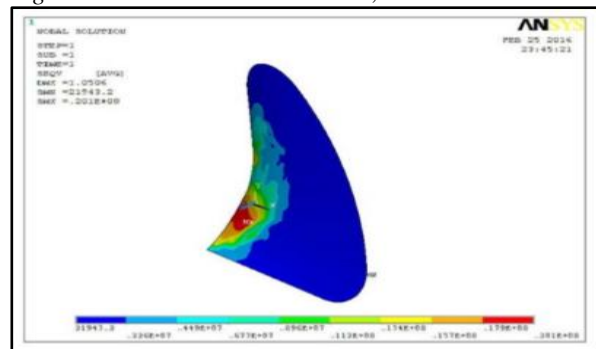


Figure 12.

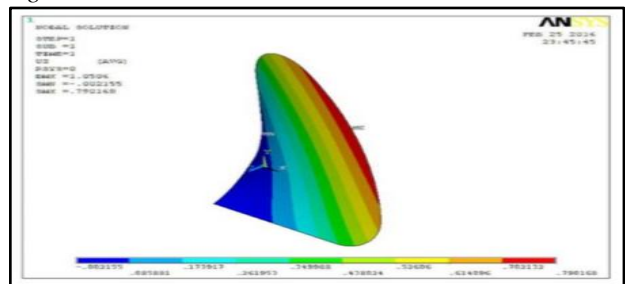


Figure 9. Deflection of CFRP blade (8 layers)

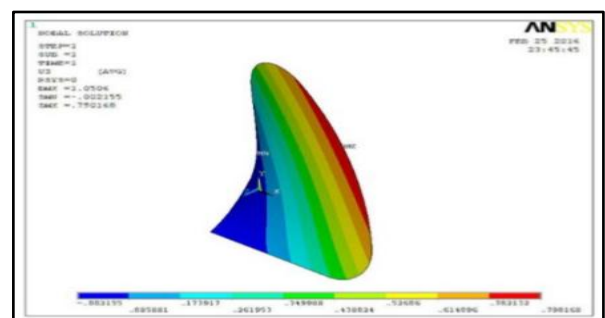


Figure 10. Deflection of hybrid composite blade (8 layers)

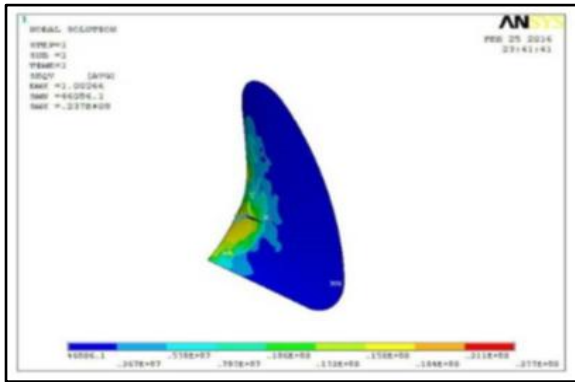


Figure 11. Max von/Mises stress for CFRP propeller blade (8 layers)

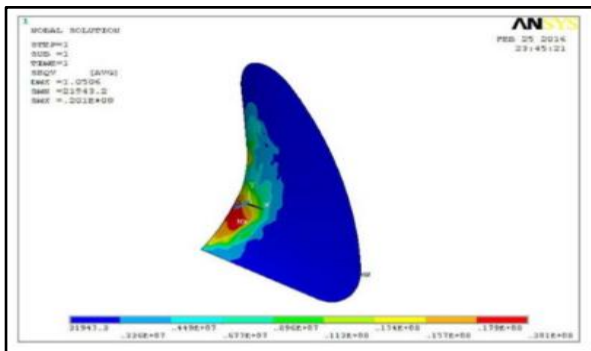


Figure 12. Max von Mises pressure/stress for hybrid composite propeller blade (8 layers)

Case 3: Analytical result for 12 coatings

Here the width of each coating is estimated as 0.0833 mm. The maximum illumination for a single CFRP propeller blade is 0.7534 mm in the direction of z i.e. normal to the blade front and the max von Mises pressure is 25.7 N / mm². In the circumstance of a hybrid propeller mounting shaft at 0.702 mm and von Mises pressure is 22.7N / mm². Both illumination & von Mises pressure/stresses are shown in Figure 13, Figure 14, Figure 15, &

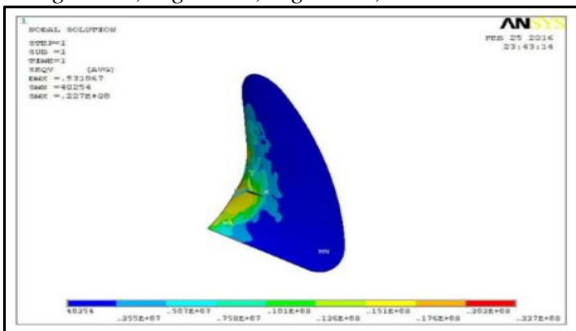


Figure 13. Deflection of CFRP blade (12 layers)

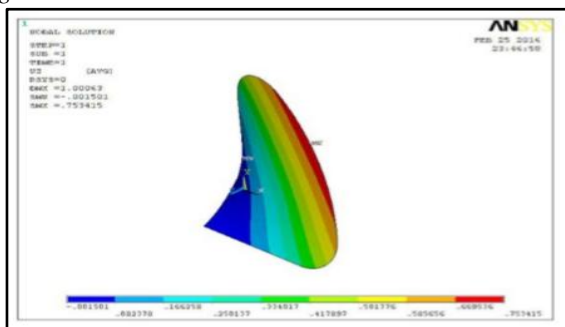


Figure 14. Deflection of hybrid composite blade (12 layers)

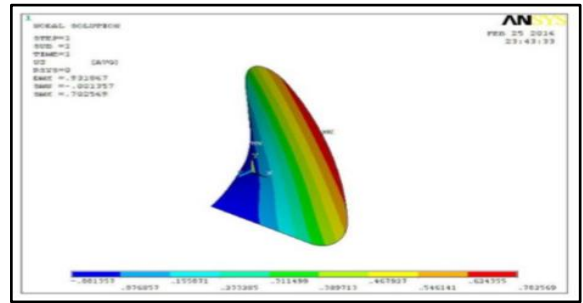


Figure 15. Max von/Mises stress for CFRP propeller blade (12 layers)

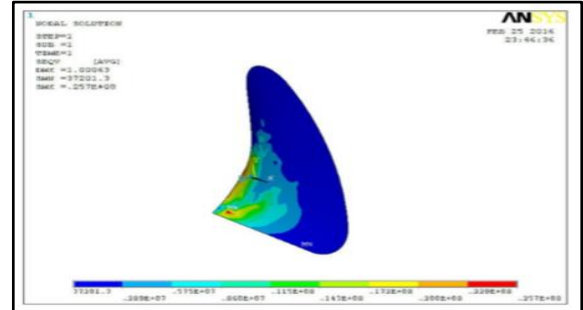


Figure 16. Max von/Mises stress for hybrid composite propeller blade (12 layers)

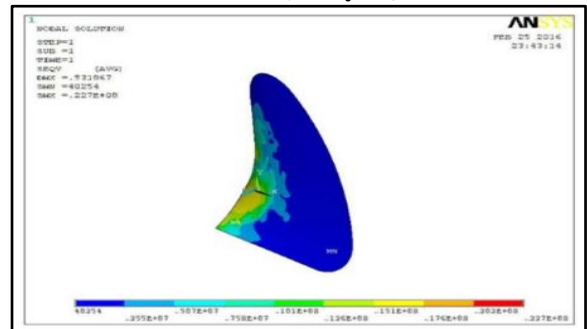


Figure 17. Deflection of CFRP blade (16 layers)

Case 4: Analytical result for 12 covers

Here the width of each coating is estimated as 0.0625 mm. The maximum illumination for a single CFRP propeller blade i.e. I normal to the blade face is 0.811 mm and the maximum pressure of von Mises is 26.3 N / mm². In the case of a hybrid propeller mounting shaft at 0.9905 mm and von Mises weight is 25.2N / mm². Both illumination and von/Mises stresses are illustrated below Figure 17, Figure 18, Figure 19, & Figure 20.

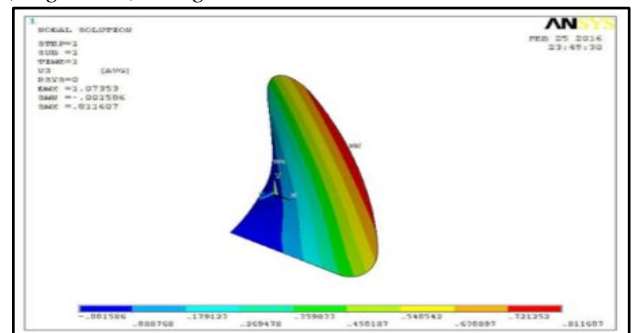


Figure 18. Deflection of CFRP blade (16 layers)

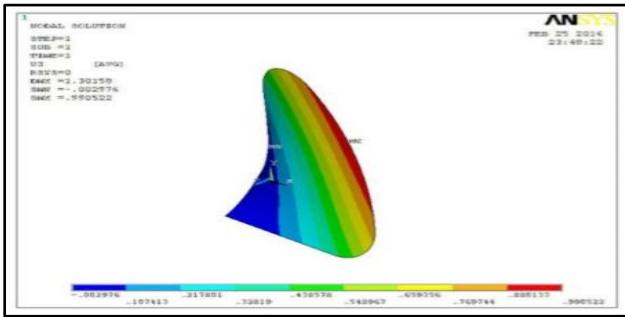


Figure 18. Deflection of hybrid composite blade (16 layers)

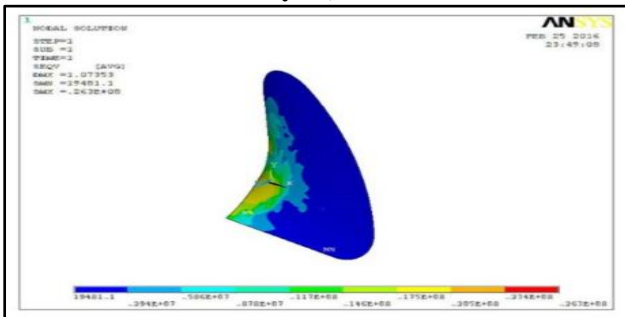


Figure 19. Max von Mises pressure/stress for CFRP propeller blade (16 layers)

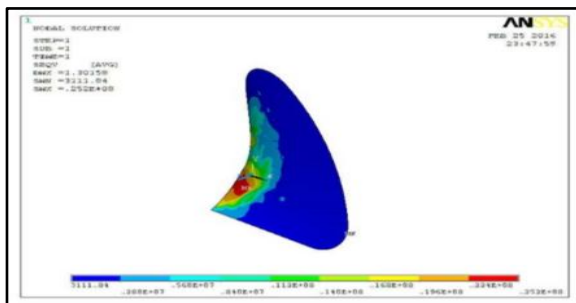


Figure 20. Max von Mises stress for hybrid composite propeller blade (16 layers)

IV. CONCLUSION

From the present work the following conclusions are drawn:

➤ The average illumination for a CFRP propeller blade is approximately 0.77925mm, where it is similar to a hybrid propeller blade at 0.85107mm. Thus a suction increase in hybrid propeller is 9.21% smaller compared to propeller compound.

The average von Mises pressure developed in CFRP is 24.45 N / mm² whereas in a hybrid propeller 22.275 N / mm². So weight reduction in a hybrid propeller blade is about 8.895%.

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